

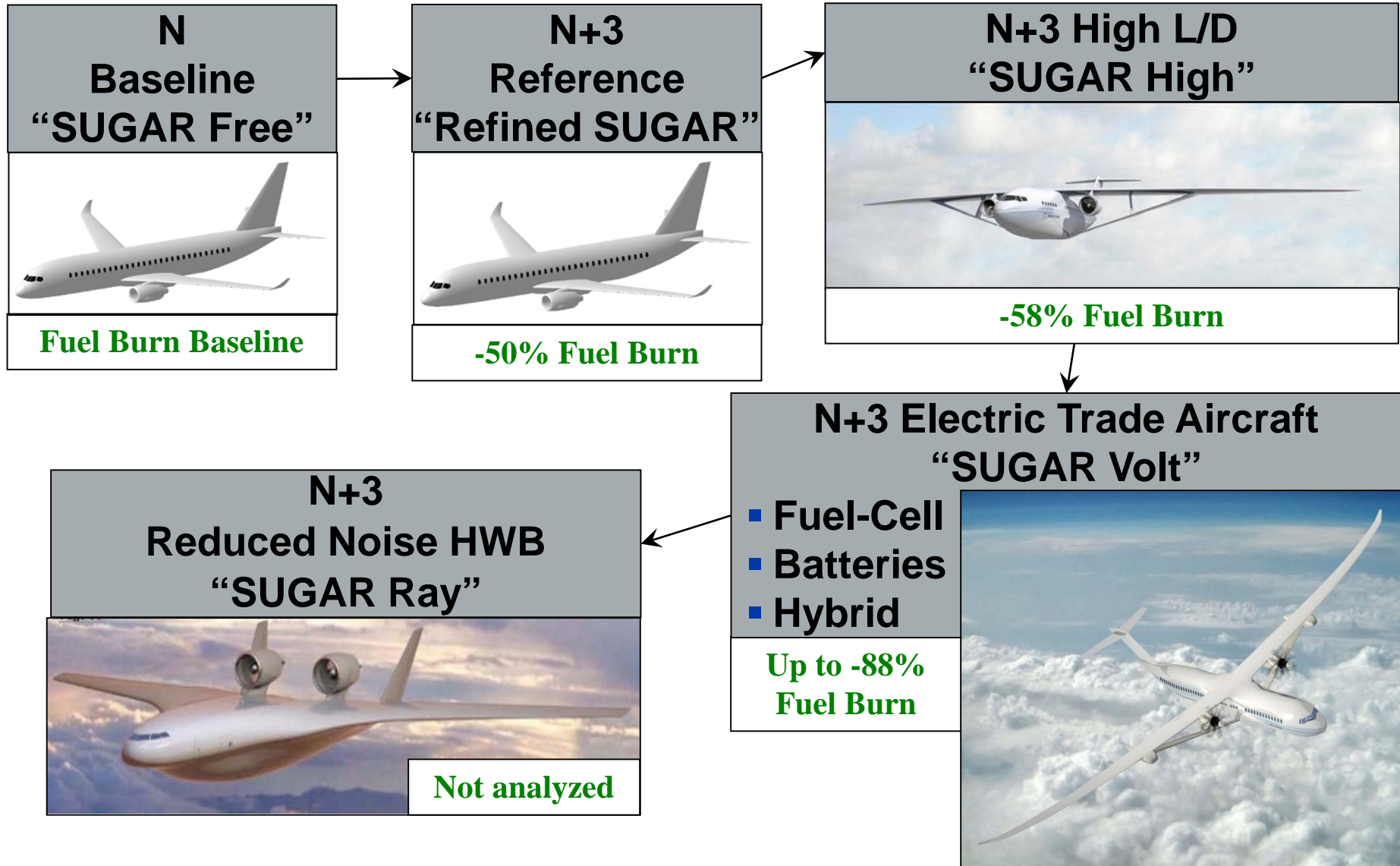


# Hybrid-Electric Propulsion

David J. Paisley  
Technical Fellow  
Product Development  
Boeing Commercial Airplanes

# Boeing N+3 Subsonic Ultra Green Airplane Research

## Concept Selections and Initial Performance

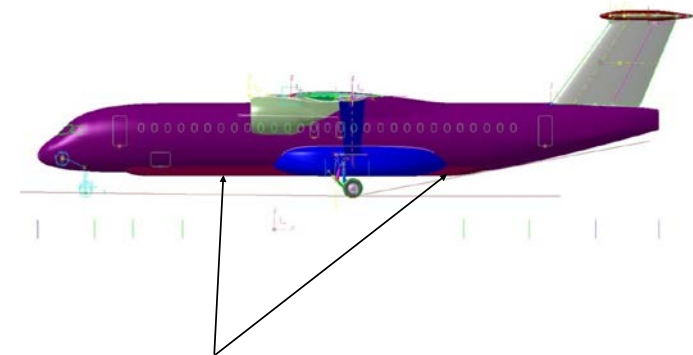


# SUGAR Volt is Derived from SUGAR High

---

- **Same as SUGAR High Except:**

- Propulsion system weight increased to 10,475 lbs
- Added battery weight dependent on range (20,900 lbs at 900 nmi for base airplane)
- Battery mounting weight 5,000 lbs
- Wire weight 1,000 lbs
- 2.5 drag counts for battery fairing
- Low wing weight version of SUGAR High wing

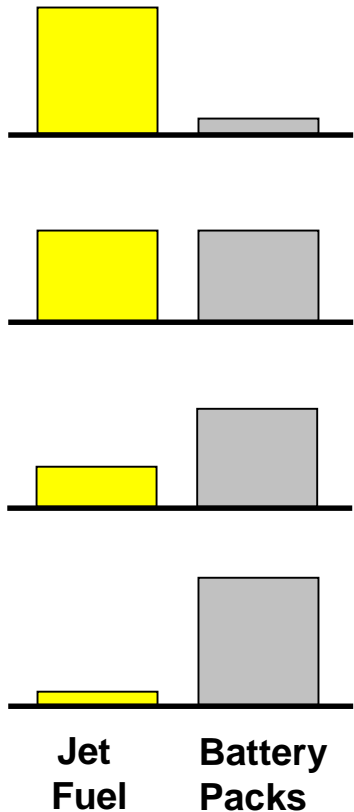


Removable Modular Battery Pack

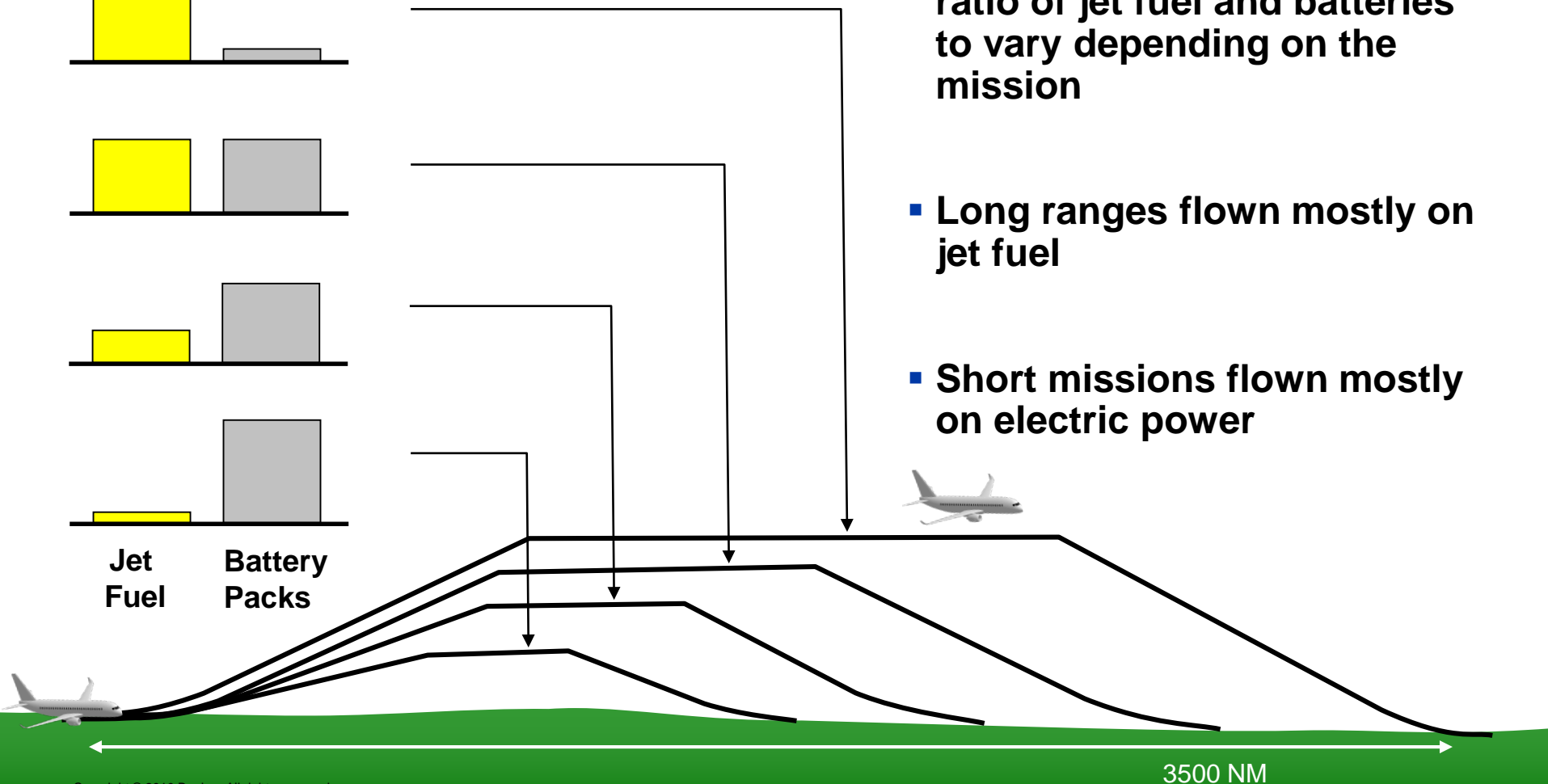
- **Note that SUGAR Volt hybrid electric engine and power system could be applied to other configurations (Refined SUGAR or SUGAR Ray)**

# Battery and Jet Fuel Loading

Weight of Jet Fuel and Batteries at takeoff



- Hybrid propulsion allows for ratio of jet fuel and batteries to vary depending on the mission
- Long ranges flown mostly on jet fuel
- Short missions flown mostly on electric power



# SUGAR Volt - Sizing

## PERFORMANCE SUMMARY SUGAR Volt

Product  
Development  
Study

200 lb / passenger  
Standard Day  
Alternate C.G. Performance

*Base SUGAR Volt achieves 63.4% fuel burn reduction*

| MODEL<br>Sizing Level                   |                      | SUGAR Free        | Refined SUGAR  | SUGAR High     | SUGAR Volt     |
|---|----------------------|-------------------|----------------|----------------|----------------|
| <b>PASSENGERS / CLASS</b>               |                      | 154 / Dual        | 154 / Dual     | 154 / Dual     | 154 / Dual     |
| <b>MAX TAKEOFF WEIGHT</b>               | LB                   | 184,800           | 139,700        | 176,800        | 154,900        |
| <b>MAX LANDING WEIGHT</b>               | LB                   | 151,000           | 131,800        | 167,300        | 148,600        |
| <b>MAX ZERO FUEL WEIGHT</b>             | LB                   | 142,000           | 123,800        | 159,300        | 140,600        |
| <b>OPERATING EMPTY WEIGHT</b>           | LB                   | 96,000            | 77,800         | 113,300        | 94,600         |
| <b>FUEL CAPACITY REQ</b>                | USG                  | 9,710             | 5,512          | 5,754          | 5,250          |
| <b>ENGINE MODEL</b>                     |                      | Scaled CFM56-7B27 | Scaled gFan    | Scaled gFan+   | Scaled hFan    |
| <b>FAN DIAMETER</b>                     | IN                   | 62                | 66             | 86             | 80             |
| <b>BOEING EQUIVLENT THRUST (BET)</b>    | LB                   | 28,200            | 15,700         | 19,600         | 17,300         |
| <b>WING AREA / SPAN</b>                 | FT <sup>2</sup> / FT | 1429 / 122        | 1440 / 129     | 1722 / 215     | 1498 / 201     |
| <b>ASPECT RATIO (EFFECTIVE)</b>         |                      | 10.41             | 11.63          | 26.94          | 26.94          |
| <b>OPTIMUM CL</b>                       |                      | 0.583             | 0.654          | 0.828          | 0.831          |
| <b>CRUISE L/D @ OPT CL</b>              |                      | 18.068            | 21.981         | 25.934         | 24.992         |
| <b>DESIGN MISSION RANGE</b>             | NMI                  | 3,500             | 3,500          | 3,500          | 3,500          |
| <b>PERFORMANCE CRUISE MACH</b>          |                      | 0.785             | 0.70           | 0.70           | 0.70           |
| <b>LONG RANGE CRUISE MACH (LRC)</b>     |                      | 0.785             | 0.70           | 0.70           | 0.70           |
| <b>THRUST ICAC (MTOW, ISA)</b>          | FT                   | 37,200            | 38,800         | 43,300         | 42,800         |
| <b>TIME / DIST (MTOW, 35k FT, ISA)</b>  | NMI / NMI            | 23 / 148          | 29 / 182       | 29 / 182       | 29 / 178       |
| <b>OPTIMUM ALTITUDE (MTOW, ISA)</b>     | FT                   | 35,000            | 38,400         | 42,100         | 42,000         |
| <b>BUFFET ICAC (MTOW, ISA)</b>          | FT                   | 36,200            | 45,200         | 44,000         | 43,900         |
| <b>TOFL (MTOW, SEA LEVEL, 86 DEG F)</b> | FT                   | 8,190             | 8,190          | 8,190          | 8,180          |
| <b>APPROACH SPEED (MLW)</b>             | KT                   | 126               | 115            | 115            | 116            |
| <b>BLOCK FUEL / SEAT (900 NMI)</b>      | LB                   | 92.35 (Base)      | 51.53 (-44.2%) | 56.43 (-38.9%) | 33.83 (-63.4%) |

40% Reduction from Sugar High

# Electric/Hybrid Electric - What is Next?

---

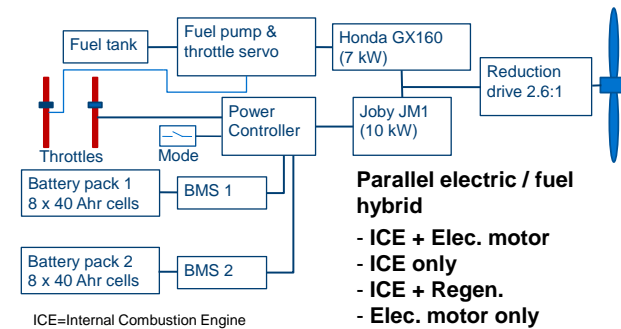
- **Everyone will keep watching battery technology development**
  - What will be the best chemistries/technologies?
  - Will trends continue progression or will they break?
- **Some will jump in and do something now!**
  - Technology demonstrators
  - Short range general aviation applications
  - Rules, regulations, and certification development
- **Battery technology and what we do now will determine the future of electric aircraft**

***Technology Not Yet Ready for Electric/Hybrid Airlines – But be Watchful!***



# Hybrid-Electric Demonstrator Aircraft

Hybrid-Electric Power System Schematic



UNIVERSITY OF CAMBRIDGE



# Advantages of Near Term Demonstrations

---

- **Design team gains and maintains needed experience**
  - Lessons learned for design and optimization
- **Public acceptance of electric aircraft**
- **Interactions with government rule makers**
- **Unexpected technical advances and improvements**
- **Creates confidence in team and technology for future investment**
- **Maintains public and industry interest**
- **Inspires students (future engineers and leaders)**
- **An opportunity to leverage resources and share knowledge across a wide range of people and organizations**



# Electric/Hybrid Electric Aircraft

---

- **Recent studies show great promise for future electric/hybrid electric aircraft**
  - Reduced environmental impact
  - Reduced energy cost & increased flexibility
  - Reduced system complexity
- **But with challenges:**
  - Dirty electric grids
  - Heavy batteries and systems
  - Safety and certification
  - Business case and technology investment
  - How do small technology demonstrations support a larger future aircraft?

